WEIGHT AND BALANCE/ EQUIPMENT LIST

TABLE OF CONTENTS

INTRODUCTION

This section describes the procedure for establishing the basic empty weight and moment of the airplane. Sample forms are provided for reference. Procedures for calculating the weight and moment for various operations are also provided. For additional information regarding Weight and Balance procedures, refer to the Aircraft Weight and Balance Handbook (FAA-H-8083-1). A comprehensive list of Cessna equipment available for this airplane is included at the back of this section.

Specific information regarding the weight, arm, moment and installed equipment for this airplane as delivered from the factory can be found in the plastic envelope in the back of this POH.

WARNING

IT IS THE RESPONSIBILITY OF THE PILOT TO MAKE SURE THE AIRPLANE IS LOADED PROPERLY. OPERATION OUTSIDE OF PRESCRIBED WEIGHT AND BALANCE LIMITATIONS COULD RESULT IN AN ACCIDENT AND SERIOUS OR FATAL INJURY.

AIRPLANE WEIGHING PROCEDURES

1. Preparation:
   a. Inflate tires to recommended operating pressures.
   b. Defuel airplane. Refer to the Maintenance Manual.
   c. Service engine oil as required to obtain a normal full indication (approximately 8 quarts on dipstick).
   d. Move sliding seats to the most forward position.
   e. Raise flaps to the fully retracted position.
   f. Place all control surfaces in neutral position.
   g. Remove all non-required items from airplane.

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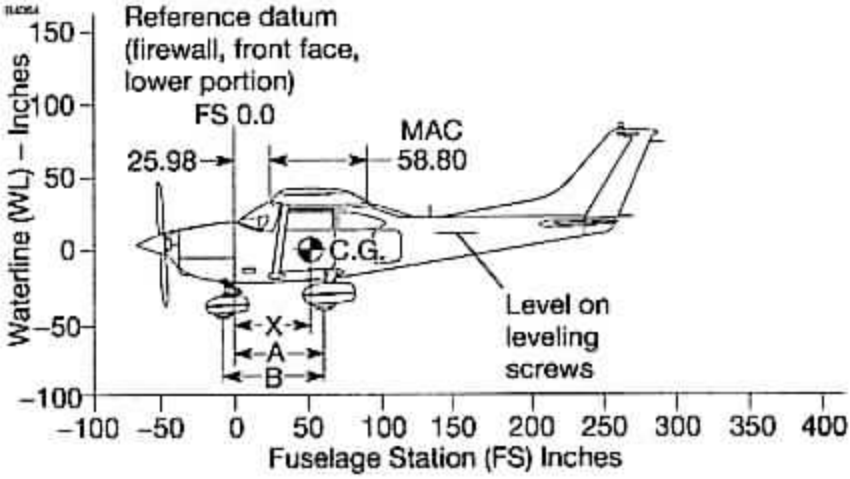
AIRPLANE WEIGHING PROCEDURES (Continued)

2. Level:
   a. Place scales under each wheel (minimum scale capacity, 1000 pounds).
   b. Deflate the nose tire and/or lower or raise the nose strut to properly center the bubble in the level (Refer to Figure 6-1).
3. Weigh:
   a. Weigh airplane in a closed hangar to avoid errors caused by air currents.
   b. With the airplane level and brakes released, record the weight shown on each scale. Deduct the tare, if any, from each reading.
4. Measure:
   a. Obtain measurement A by measuring horizontally (along the airplane centerline) from a line stretched between the main wheel centers to a plumb bob dropped from the firewall.
   b. Obtain measurement B by measuring horizontally and parallel to the airplane centerline, from center of nosewheel axle, left side, to a plumb bob dropped from the line between the main wheel centers. Repeat on right side and average the measurements.
5. Using weights from step 3 and measurements from step 4, the Basic Empty Weight and C.G. can be determined by completing Figure 6-1 (Sheet 2).
6. Changes to the Airplane Weight and Balance due to alteration or repair must be documented in a permanent record within the POH similar to that shown in Figure 6-2.
7. A new Basic Empty Weight and CG Arm based on actual airplane weight (as weighed) is required after a major repair or alteration. It is recommended that the airplane be weighed to verify Basic Empty Weight and CG Arm at intervals not to exceed 5 years.

AIRPLANE WEIGHING FORM



NOTE

It is the responsibility of the pilot to make sure that the airplane is loaded properly.

Figure 6-1 (Sheet 1 of 2)

AIRPLANE WEIGHING FORM

Locating CG with Airplane on Landing Gear

$$X \text{ (Inches Aft of Datum)} = A - \left[\frac{\text{Nosewheel Weight} \times B}{\text{Total Weight*}}\right]$$

*(Nose + L + R Wheel Weights)

Locating Percent MAC

$$\text{CG Percent MAC} = \frac{(\text{CG Arm of Airplane}) - 25.98}{0.5880}$$

Leveling Provisions

Longitudinal – Left side of tailcone at FS 139.65 and 171.65

Measuring A and B

Measure A and B per pilot's operating handbook instructions to assist in locating CG with airplane weighed on landing gear.

Airplane as Weighed Table

Position	Scale reading	Scale drift	Tare	Net weight
Left Wheel				
Right Wheel				
Nose Wheel				
Airplane total as weighed				

Basic Empty Weight and Center-of-Gravity Table

Item	Weight Pounds	CG Arm (Inches)	Moment (Inch-Pounds /1000)
Airplane (calculated or as weighed) (includes all undrainable fluids and full oil)			
Drainable unusable fuel at 6.0 pounds per gallon – (5 gallons)	30.0	48.00	1.4
Basic Empty Weight			

Figure 6-1 (Sheet 2)

SAMPLE WEIGHT AND BALANCE RECORD

03009

SAMPLE WEIGHT AND BALANCE RECORD

(Continuous history of changes in structure or equipment affecting weight and balance)

Date	Item no. In	Item no. Out	Airplane model	Description of article or modification	Serial number	Page number	Weight change Added (+) WT. (lb.)	Added (+) Arm (in.)	Added (+) Moment /1000	Removed (-) WT. (lb.)	Removed (-) Arm (in.)	Removed (-) Moment /1000	Running basic empty weight WT. (lb.)	Running basic empty weight Moment /1000
				As delivered										

0585T1009

Figure 6-2

WEIGHT AND BALANCE

The following information will enable you to operate your Cessna within the prescribed weight and center of gravity limitations. To determine weight and balance, use the Sample Loading Problem (Figure 6-3), Loading Graph (Figure 6-4), and Center of Gravity Moment Envelope (Figure 6-7) as follows:

Enter the appropriate basic empty weight and moment/1000 from the weight and balance records for your airplane in the YOUR AIRPLANE column of the Sample Loading Problem.

NOTE

In addition to the basic empty weight and moment noted on these records, the C.G. arm (FS) is also shown, but need not be used on the Sample Loading Problem. The moment which is shown must be divided by 1000 and this value used as the moment/1000 on the loading problem.

Use the Loading Graph to determine the moment/1000 for each additional item to be carried; then list these on the loading problem.

NOTE

Loading Graph information for the pilot, passengers and baggage is based on seats positioned for average occupants and baggage loaded in the center of the baggage areas as shown on the Loading Arrangements diagram. For loadings which may differ from these, the Sample Loading Problem lists fuselage stations (FS) for these items to indicate their forward and aft C.G. range limitations (seat travel and baggage area limitation). Refer to Figures 6-5 and 6-6 for additional loading information. Additional moment calculations, based on the actual weight and C.G. arm (FS) of the item being loaded, must be made if the position of the load is different from that shown on the Loading Graph.

Total the weights and moments/1000 and plot these values on the Center of Gravity Moment Envelope to determine whether the point falls within the envelope, and if the loading is acceptable.

(Continued Next Page)

WEIGHT AND BALANCE (Continued)

BAGGAGE TIEDOWN

A nylon baggage net having four tiedown straps is provided as standard equipment to secure baggage in the area aft of the rear seat (baggage areas, A, B and C). Eight eyebolts serve as attaching points for the net. A placard on the baggage door defines the weight limitations in the baggage areas.

When baggage area A is utilized for baggage only, the four forward eyebolts should be used. When only baggage area B is used, the eyebolts just aft of the baggage door and the eyebolts above or below the shelf area may be used. When only baggage area C is utilized, the eyebolts above and below the shelf area should be used. When the cabin floor (baggage areas A and B) is utilized for baggage, the four forward eyebolts and the eyebolts mounted above or below the shelf area should be used. When there is baggage in areas B and C, the eyebolts just aft of the baggage door and the eyebolts above and below the shelf area should be used.

(Continued Next Page)

WEIGHT AND BALANCE (Continued)

BAGGAGE TIEDOWN (Continued)

When baggage is contained in all three areas, the two forward eyebolts on the cabin floor, the eyebolts just aft of the baggage door or the eyebolts at the bottom of the forward portion of the shelf area and the eyebolts near the upper forward surface of the shelf area should be used.

The rear bench seat can be removed to access the floorboard area of the rear cabin. Baggage may then be tied down using ten tiedown eyebolts to standard attach points located in the interior area of the airplane shown in Figure 6-6.

The maximum allowable floor loading of the rear cabin area is 200 pounds/square foot; however, when items with small or sharp support areas are carried, the installation of a plywood floor is recommended to protect the airplane structure.

The maximum rated load weight capacity for each of the ten tiedowns is 140 pounds. Rope, strap or cable used for tiedown should be rated at a minimum of ten times the load weight capacity of the tiedown fittings used. Weight and balance calculations for items in the area of the rear seat and baggage area can be figured on the Loading Graph using the lines labeled Rear Passengers or Cargo.

SAMPLE LOADING PROBLEM

ITEM DESCRIPTION	WEIGHT AND MOMENT TABULATION			
	SAMPLE AIRPLANE		YOUR AIRPLANE	
	Weight (lbs)	Moment (lb-ins/ 1000)	Weight (lbs)	Moment (lb-ins/ 1000)
1 - Basic Empty Weight (Use the data pertaining to your airplane as it is presently equipped. Includes unusable fuel and full oil)	1924	70.9		
2 - Usable Fuel (At 6 Lbs./Gal.)				
- Standard Fuel - 87 Gallons Maximum	522	24.3		
- Reduced Fuel - 64 Gallons				
3 - Pilot and Front Passenger (FS 32 to 50)	340	12.6		
4 - Rear Passengers (FS 74)	200	14.8		
Cargo - Replacing Rear Passenger Seat (FS 65 to 82)				
5 - *Baggage "A" (FS 82 to 109) 120 Pounds Maximum	100	9.7		
*Baggage "B" (FS 109 to 124) 80 Pounds Maximum	24	2.8		
*Baggage "C" (FS 124 to 134) 80 Pounds Maximum				
6 - RAMP WEIGHT AND MOMENT	**3110**	**135.1**		
7 - Fuel allowance for engine start, taxi and runup	-10.0	-0.5		
8 - **TAKEOFF WEIGHT AND MOMENT** (Subtract Step 7 from Step 6)	**3100**	**134.6**		

9 - Locate this point (3100 at 134.6) on the Center of Gravity Moment Envelope, and since this point falls within the envelope, the loading is acceptable, providing that flight time is allowed for fuel burn-off to a maximum of 2950 pounds before landing.

*The maximum allowable combined weight capacity for baggage in areas A, B and C is 200 pounds. The maximum allowable combined weight capacity in areas B and C is 80 pounds.

Figure 6-3 (Sheet 1 of 2)

SAMPLE LOADING PROBLEM

YOUR AIRPLANE	
Weight (lbs.)	Moment (Lb-ins. /1000)

YOUR AIRPLANE	
Weight (lbs.)	Moment (Lb-ins. /1000)

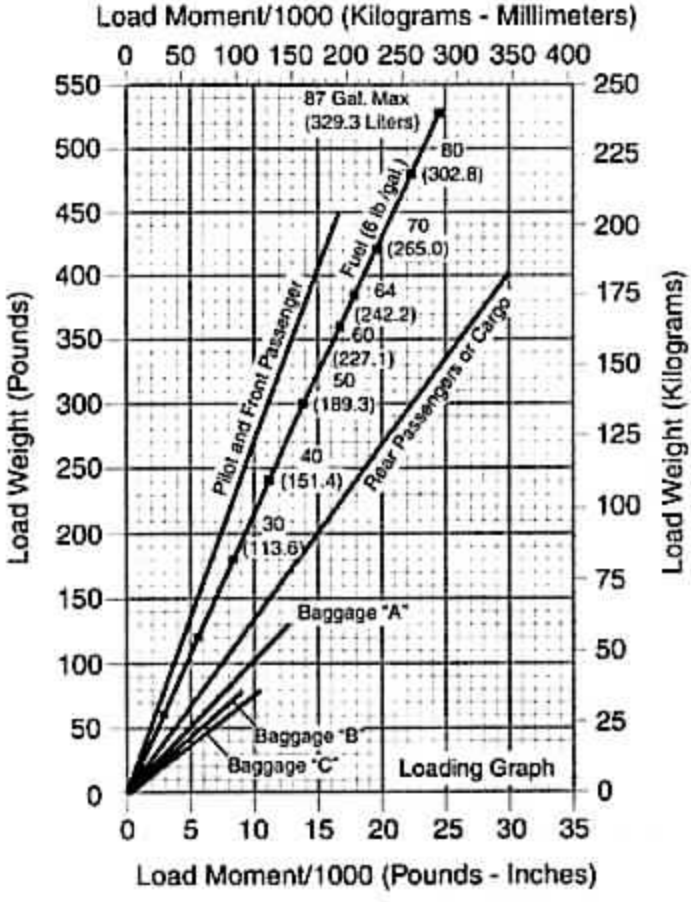
YOUR AIRPLANE	
Weight (lbs.)	Moment (Lb-ins. /1000)

NOTE

When several loading configurations are representative of your operations, it may be useful to fill out one or more of the above columns so specific loadings are available at a glance.

Figure 6-3 (Sheet 2)

LOADING GRAPH

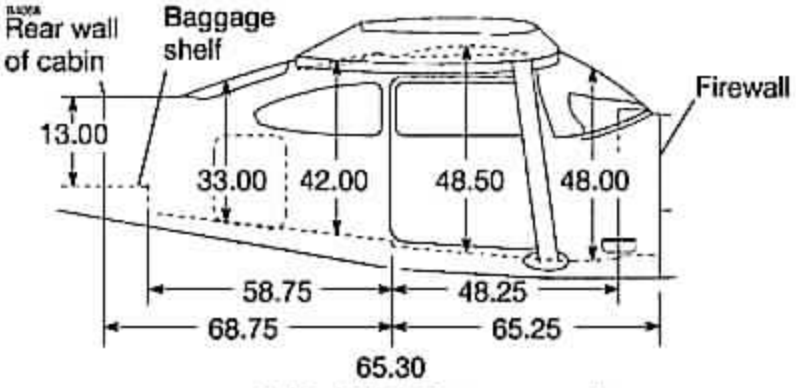


NOTE

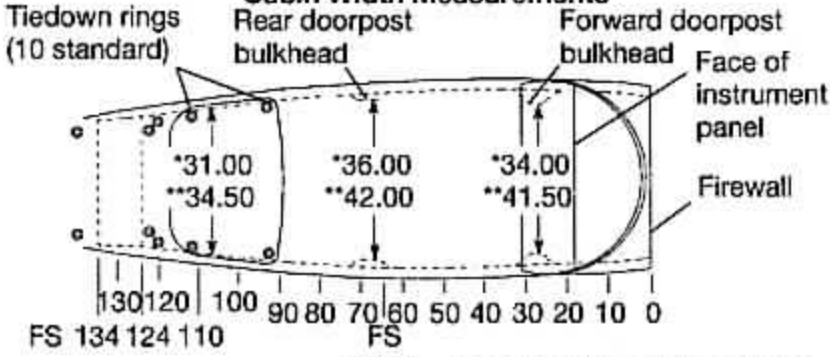
Line representing adjustable seats shows the pilot and front seat passenger center of gravity on adjustable seats positioned for average occupant. Refer to the Loading Arrangements diagram for forward and aft limits of occupant C.G. range.

Figure 6-4

INTERNAL CABIN DIMENSIONS



Cabin Width Measurements



CODE

*Cabin floor
**Lower window line

Door Opening Dimensions

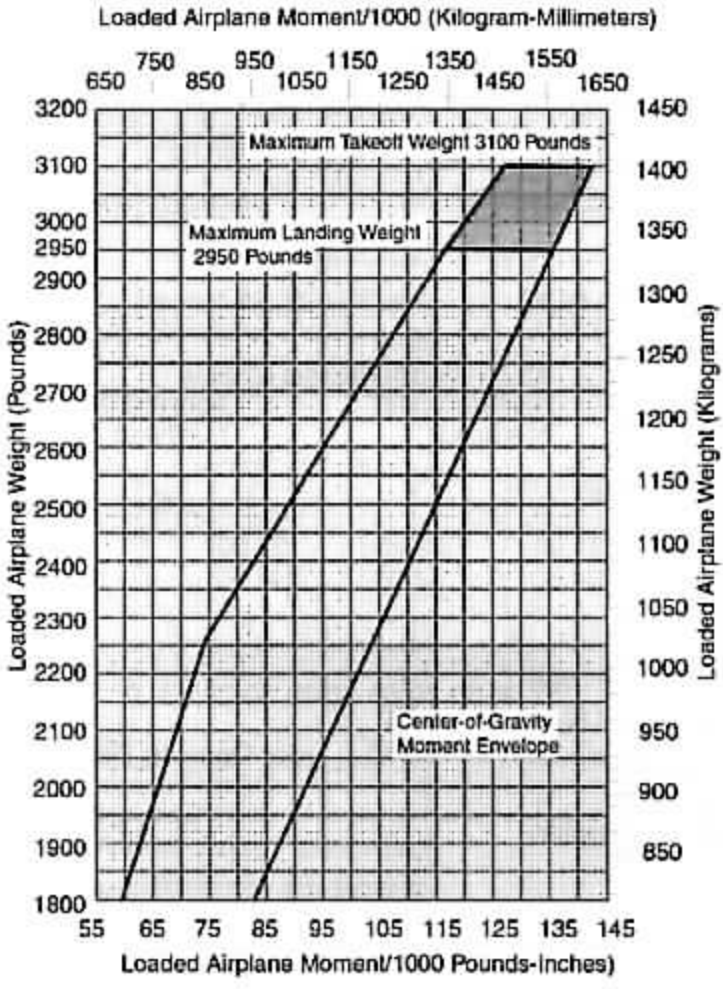
	Width (top)	Width (bottom)	Height (front)	Height (rear)
Cabin doors	32.00	36.50	41.00	38.50
Baggage door	15.75	15.75	22.00	20.50

NOTE

- Maximum allowable floor loading is 200 pounds per square foot.
- All dimensions shown are in inches.

Figure 6-6

CENTER OF GRAVITY MOMENT ENVELOPE

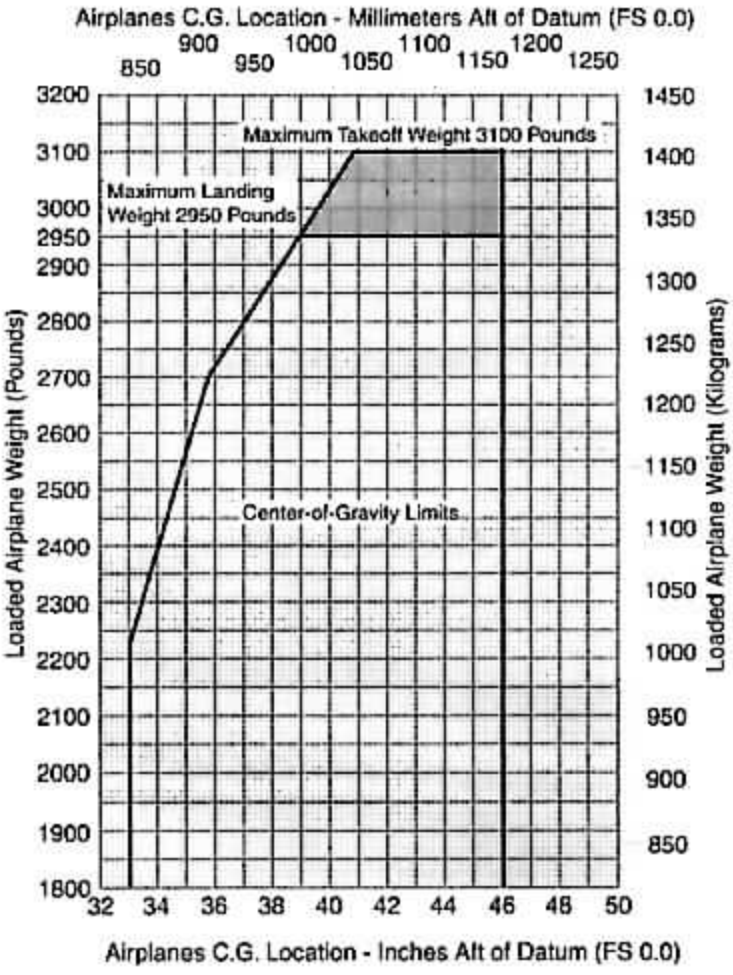


NOTE

If takeoff weight is more than maximum landing weight, allow flight time for fuel burn off to 2950 pounds before landing.

Figure 6-7

CENTER OF GRAVITY LIMITS



NOTE

If takeoff weight is more than maximum landing weight, allow flight time for fuel burn off to 2950 pounds before landing.

Figure 6-8

COMPREHENSIVE EQUIPMENT LIST

Figure 6-9 is a comprehensive list of all Cessna equipment which is available for the Model 182T airplane equipped with Garmin G1000 Integrated Cockpit System and GFC 700 Autopilot. This comprehensive equipment list provides the following information in column form:

In the **ITEM NO** column, each item is assigned a coded number. The first two digits of the code represent the identification of the item within Air Transport Association Specification 100 (11 for Paint and Placards; 24 for Electrical Power; 77 for Engine Indicating, etc.). These assignments also correspond to the Maintenance Manual chapter for the airplane. After the first two digits, items receive a unique sequence number (01, 02, 03, etc.). After the sequence number, a suffix letter is assigned to identify equipment as a required item, a standard item or an optional item.

Suffix letters are as follows:

R = Required items or equipment for FAA certification (14 CFR 23 or 14 CFR 91).

S = Standard equipment items.

O = Optional equipment items replacing required or standard items.

A = Optional equipment items which are in addition to required or standard items.

In the **EQUIPMENT LIST DESCRIPTION** column, each item is assigned a descriptive name to help identify its function.

In the **REF DRAWING** column, a Cessna drawing number is provided which corresponds to the item.

NOTE

If additional equipment is to be installed, it must be done in accordance with the reference drawing, service bulletin or a separate FAA approval.

In the **WT LBS** and **ARM INS** columns, information is provided on the weight (in pounds) and arm (in inches) of the equipment item.

NOTE

- Unless otherwise indicated, true values (not net change values) for the weight and arm are shown. Positive arms are distances aft of the airplane datum; negative arms are distances forward of the datum.
- Asterisks (*) in the weight and arm column indicate complete assembly installations. Some major components of the assembly are listed on the lines immediately following. The sum of these major components does not necessarily equal the complete assembly installation.

ITEM NO	EQUIPMENT LIST DESCRIPTION	REF DRAWING	WT LBS	ARM INS.
	11 - PAINT AND PLACARDS			
11-01-S	PAINT, OVERALL WHITE WITH COLOR STRIPE	0700810	19.6*	92.9*
	- OVERALL WHITE COLOR		18.8	91.5
	- COLOR STRIPING		0.8	135.9
	21 - AIR CONDITIONING			
21-01-S	VENTILATORS, ADJUSTABLE, CABIN AIR	0719010	1.7	38.5
21-02-S	CABIN HEATER SYSTEM, SHROUDED MUFFLER TYPE	0750635	2.5	-29.5
21-03-R	FORWARD AVIONICS COOLING FAN - MC24B3	3930370	0.5	12.7
21-04-R	AFT AVIONICS COOLING FAN	3940389	1.1	125.5
	22 - AUTO FLIGHT			
22-01-S	GFC 700 AUTOPILOT		6.9*	141.1*
	- PITCH SERVO	3940452-1	2.3	185.9
	- PITCH TRIM SERVO	3940454-1	2.3	176.4
	- ROLL SERVO	3940453-1	2.3	61.0
	23 - COMMUNICATIONS			
23-01-S	STATIC DISCHARGE WICKS, (SET OF 11)	1201131-2	0.3	152.9
23-02-R	AUDIO/INTERCOM/MARKER BEACON	3930368		
	- GMA 1347 AUDIO PANEL	3910317-2	2.6	16.5
	- CI-102 MARKER BEACON ANTENNA	3960193-2	0.5	131.5
23-03-R	NAV/COM/GPS #1 COMPUTER	3940389		
	- GIA 63W INTEGRATED AVIONICS UNIT	3910317-4	4.9	134.0
	- CI 2580-200 VHF COMM/GPS ANTENNA	3960222-7	0.5	61.2
23-04-S	NAV/COM/GPS #2 COMPUTER	3940389		
	- GIA 63W INTEGRATED AVIONICS UNIT	3910317-4	4.9	134.0
	- CI 2580-410 VHF COMM/GPS/XM ANTENNA	3960222-8	0.5	61.2
	or			
	- CI 2580-200 VHF COMM/GPS ANTENNA	3960222-9	0.5	61.2
	- CI 420-10 XM ANTENNA	3960234-1	0.5	41.5
	24 - ELECTRICAL POWER			
24-01-R	ALTERNATOR, 28 VOLT, 60 AMP, -9910591-11	0750635	10.0	-33.4
24-02-O	ALTERNATOR, 28 VOLT, 95 AMP, -9910592-3	0750636	15.7	-33.4
24-03-R	BATTERY, 24 VOLT, 8.00 AMP HOUR	0718012	23.2	132.1
24-04-O	BATTERY, 24 VOLT, 10.00 AMP HOUR	0701169	27.2	132.1
24-05-R	POWER DISTRIBUTION MODULE S3100-366		6.4*	-2.5*
	- ALTERNATOR CONTROL UNIT	AC2101	0.2	-2.5
	- MASTER CONTACTOR	X61-0007	0.7	-2.5
	- STARTER CONTACTOR	X61-0027	0.7	-2.5
	- AMMETER TRANSDUCER	CS3200	0.1	-2.0
24-06-S	BATTERY, STANDBY - AVT 200413, 24 VOLT, 6.20 AMP HOUR	0718023-1	14.0	10.8

Figure 6-9 (Sheet 1 of 6)

ITEM NO	EQUIPMENT LIST DESCRIPTION	REF DRAWING	WT LBS	ARM INS.
	25 - EQUIPMENT/FURNISHINGS			
25-01-R	SEAT, PILOT, ADJUSTABLE, CLOTH/VINYL COVER	0719113-1	33.8	41.5
25-02-O	SEAT, PILOT, ADJUSTABLE, LEATHER/VINYL COVER	0719114-1	34.3	41.5
25-03-S	SEAT, FRONT PASSENGER, ADJUSTABLE, CLOTH/VINYL COVER	0719113-2	33.8	41.5
25-04-O	SEAT, FRONT PASSENGER, ADJUSTABLE, LEATHER/VINYL COVER	0719114-2	34.3	41.5
25-05-S	SEAT, REAR PASSENGER, TWO PIECE BACK, CLOTH/VINYL COVER	0719115-1	50.0	82.0
25-06-O	SEAT, REAR PASSENGER, TWO PIECE BACK, LEATHER/VINYL COVER	0719116-1	51.0	82.0
25-07-R	SEAT BELT AND SHOULDER HARNESS, INERTIA REEL, AUTO ADJUST, PILOT AND FRONT PASSENGER	0719087	5.2	50.3
25-08-S	SEAT BELT AND SHOULDER HARNESS, INERTIA REEL, AUTO ADJUST, REAR SEAT	0719087	5.2	87.8
25-09-S	SUN VISOR (SET OF 2)	0514166-3	1.2	33.0
25-10-S	BAGGAGE RESTRAINT NET	1215171-2	0.5	109.0
25-11-S	CARGO TIEDOWN RINGS (SET OF 10)	FD-2997M34-2	0.4	108.0
25-12-S	TOW BAR, NOSE GEAR (STOWED)	0501019-1	1.7	108.0
25-13-R	PILOT'S OPERATING HANDBOOK AND FAA APPROVED AIRPLANE FLIGHT MANUAL (STOWED IN FRONT PASSENGER'S SEAT BACK)	0700765-2	2.2	49.5
25-14-R	GARMIN G1000 COCKPIT REFERENCE GUIDE (STOWED IN COCKPIT SIDE PANEL POCKET)		1.5	15.0
25-15-O	APPROACH PLATE HOLDER	2619012-1	0.1	22.0
25-16-S	FUEL SAMPLING CUP (STOWED)	0500838-1	0.1	49.5
25-17-S	ARTEX ME406 - 2 FREQUENCY ELT	3940461-1	2.6*	137.7*
	- ELT TRANSMITTER	ME406	2.1	134.2
	- ANTENNA AND CABLE ASSY	110-338	0.5	152.4
25-18-O	ARTEX C406-N - 3 FREQUENCY ELT	3940462-1	5.1*	139.1*
	- ELT TRANSMITTER	C406-N	4.6	137.7
	- ANTENNA AND CABLE ASSY	110-338	0.5	152.4

Figure 6-9 (Sheet 2)

ITEM NO	EQUIPMENT LIST DESCRIPTION	REF DRAWING	WT LBS	ARM INS.
	26 - FIRE PROTECTION			
26-01-S	FIRE EXTINGUISHER	0501011-3	5.3*	29.0*
	- FIRE EXTINGUISHER, HAND TYPE	A352GS	4.8	29.0
	- MOUNTING CLAMP AND HARDWARE	1290010-1	0.5	29.0
	27 - FLIGHT CONTROLS			
27-01-S	DUAL CONTROLS, RIGHT SEAT	0706015-1	5.9*	12.9*
	- CONTROL WHEEL, COPILOT	0713377-4	2.3	26.0
	- RUDDER AND BRAKE PEDAL, COPILOT	0760650-4	3.6	6.8
27-02-O	RUDDER PEDAL EXTENSION (SET OF 2) (INSTALLED ARM SHOWN)	0501082-2	2.9	8.0
	28 - FUEL			
28-01-R	AUXILIARY FUEL PUMP - 5100-00-4	0716158	1.9	-12.0
28-02-R	FUEL SENDER, FLOAT TYPE - S3852-3, -4	0720701	0.1	56.3
	or			
	FUEL SENDER, VIBROMETER - 76-207-4, -5	0720701	0.9	56.3
	30 - ICE AND RAIN PROTECTION			
30-02-S	STALL SENSOR HEAT AND PITOT HEAT	0720701	0.7	28.0
	31 - INDICATING/RECORDING SYSTEM			
31-01-S	RECORDING HOURMETER - C664503-0103	0706015	0.5	16.7
31-02-R	STALL WARNING SYSTEM			
	- STALL WARNING HORN - 0718007-1	0718015	0.5	40.0
	- WING UNIT, STALL WARNING - S1672-9	0720701	0.3	25.6
31-03-R	GEA 71 ENGINE/AIRFRAME UNIT	3930368	2.2	11.4
31-04-R	GTP 59 OUTSIDE AIR TEMPERATURE (OAT) PROBE	0706015	0.1	41.5
	32 - LANDING GEAR			
32-01-R	WHEEL BRAKE AND TIRE, 6.00 X 6 MAIN (2)	0741625-21	37.1*	58.6*
	- WHEEL ASSY CLEVELAND 40-75B (EACH)	C163001-0301	7.8	58.9
	- BRAKE ASSY CLEVELAND 30-52 (EACH)	030-05219-1	1.8	55.5
	- TIRE, 6-PLY (EACH)	C262003-0204	7.9	58.9
	- TUBE (EACH)	C262023-0102	1.3	58.9
32-02-R	WHEEL AND TIRE ASSY, 5.00 X 5 NOSE	0540000-2	6.8*	-7.1*
	- WHEEL ASSY CLEVELAND 40-77	1241156-12	2.8	-7.1
	- TIRE, 6-PLY	C262003-0202	4.6	-7.1
	- TUBE	C262023-0101	1.4	-7.1
32-03-A	WHEEL FAIRING AND INSTALLATION	0741650-1	16.4*	44.5*
	- WHEEL FAIRING, NOSE	0742411-1	3.1	-6.0
	- WHEEL FAIRINGS, MAIN (SET OF 2)	0741648-1, -2	9.5	60.6
32-04-A	HUB CAPS, WHEELS	0741048-8	0.1	62.1

Figure 6-9 (Sheet 3)

ITEM NO	EQUIPMENT LIST DESCRIPTION	REF DRAWING	WT LBS	ARM INS.
	33 - LIGHTS			
33-01-S	MAP LIGHT IN CONTROL WHEEL	0760149-5	0.2	21.5
33-02-S	COURTESY LIGHTS UNDER WING	0700615-16	0.7	61.7
33-03-S	FLASHING BEACON, GROUND RECOGNITION	0701042-6, -7	0.8	253.1
33-04-R	STROBE LIGHT	0723207	3.2	40.4
33-05-S	LANDING AND TAXI LIGHT, HID	1221059-14, -15	2.2	26.8
	34 - NAVIGATION			
34-01-R	STANDBY AIRSPEED INDICATOR - S3325-9	0706015	0.7	16.2
34-02-R	STANDBY ATTITUDE INDICATOR - S3326-8	0706015	2.2	14.0
34-03-R	STANDBY ALTIMETER, SENSITIVE WITH 20 FOOT MARKINGS, INCHES OF MERCURY AND MILLBARS - S3827-1	0706015	0.9	15.3
34-04-S	ALTERNATE STATIC AIR SOURCE	0701028-4	0.2	15.5
34-05-R	COMPASS, MAGNETIC	1213679-5	0.5	18.0
34-06-R	TRANSPONDER	3940389		
	- GTX-33 TRANSPONDER	3910317-5	3.6	134.0
	- CI 105-16 TRANSPONDER ANTENNA	3960195	0.4	86.5
34-07-R	PFD DISPLAY	3930374		
	- GDU-1044B DISPLAY	3910317-1	6.5	15.0
34-08-R	MFD DISPLAY	3930368		
	- GDU-1044B DISPLAY	3910317-1	6.5	15.0
34-09-R	ATTITUDE HEADING REFERENCE SENSOR (AHRS)	3940389		
	- GRS 77 AHRS	3910317-3	2.6	134.0
	- GMU 44 MAGNETOMETER	3940394	0.3	44.0
34-10-R	AIR DATA COMPUTER	3930368		
	- GDC 74A AIR DATA COMPUTER	3910317-6	3.6	11.4
34-11-O	WX 500 STORMSCOPE			
	- PROCESSOR - S3100-276	3940403	2.5	162.0
	- ANTENNA - 805-10930-001	3960206	1.0	175.4
34-12-S	GDL-69A DATALINK	3930368	2.8	11.4
34-13-O	AUTOMATIC DIRECTION FINDER (ADF)			
	- KR 87 ADF RECEIVER	3930495	3.2	12.2
	- ADF ANTENNA	3960192	4.2	39.3
34-14-O	DISTANCE MEASURING EQUIPMENT (DME)			
	- KN 63 REMOTE DME	3940449	2.8	177.2
	- CI 105-16 DME ANTENNA	3960232	0.4	114.5
34-15-O	KTA 810 TRAFFIC ADVISORY SYSTEM	3940441	8.8	165.0

Figure 6-9 (Sheet 4)

ITEM NO	EQUIPMENT LIST DESCRIPTION	REF DRAWING	WT LBS	ARM INS.
	37 - VACUUM			
37-01-R	ENGINE DRIVEN VACUUM PUMP			
	- VACUUM PUMP - AA3215CC	0706009	2.1	-5.0
	- COOLING SHROUD	1201998-1	0.2	-5.6
	- FILTER	1201075-2	0.3	11.5
	- VACUUM REGULATOR	AA2H3-2	0.5	2.1
37-02-R	VACUUM TRANSDUCER - P165-5786	0706015	0.3	8.5
	53 - FUSELAGE			
53-01-S	REFUELING STEPS AND HANDLE	0701127-2	1.7	15.2
53-02-A	STABILIZER ABRASION BOOTS (SET OF 2)	0500041-3	0.6	206.0
	56 - WINDOWS			
56-01-S	WINDOW, HINGED RIGHT SIDE	0711050-50	3.9	48.0
56-02-S	WINDOW, HINGED LEFT SIDE	0711050-49	3.9	48.0
	61 - PROPELLER			
61-01-R	PROPELLER ASSEMBLY, 3-BLADE OIL FILLED HUB - P4317296-01 MCCAULEY B3D36C431/80VSA-1 (WEIGHT WITHOUT 30-01-S)	0750630	76.6	-47.5
61-02-R	- SPINNER, 3-BLADE - D-7261-1	0750630	4.1	-49.9
61-03-R	- GOVERNOR, PROPELLER - C161031-0119	0750635	2.7	-42.5
	71 - POWERPLANT			
71-01-R	FILTER, INDUCTION AIR - P106150	0750635	1.3	-35.2
71-02-O	WINTERIZATION KIT (INSTALLED ARM SHOWN)	0752733-12	1.2	-42.0
	72 - ENGINES			
72-01-R	ENGINE, LYCOMING IO-540-AB1A5	0750635-1	400.4*	-23.6*

Figure 6-9 (Sheet 5)

ITEM NO	EQUIPMENT LIST DESCRIPTION	REF DRAWING	WT LBS	ARM INS.
	73 - ENGINE FUEL AND CONTROL			
73-01-R	MANIFOLD PRESSURE TRANSDUCER - P165-30A-E4C	0706015	0.1	-8.5
73-02-R	FUEL FLOW TRANDUCER - 680501K	0701171	0.8	-12.4
	77 - ENGINE INDICATING			
77-01-R	ENGINE TACHOMETER SENSOR - 1A3C-2	0701171	0.2	-8.0
77-02-R	CYLINDER HEAD THERMOCOUPLES (ALL CYLINDERS) - 32DKWUE006F0126	0701171	0.2	-12.0
77-03-S	EXHAUST THERMOCOUPLES (ALL CYLINDERS) - 86317	0701171	0.3	-12.0
	78 - EXHAUST			
78-01-R	EXHAUST SYSTEM	0750635		
	- EXHAUST SYSTEM - LEFT	9954200-13	8.4	-24.2
	- EXHAUST SYSTEM - RIGHT	9954200-14	8.4	-24.2
	79 - OIL			
79-01-R	OIL COOLER - 10610R	0750635	5.5	-11.4
79-02-R	OIL PRESSURE SENSOR - P165-5281	0750635	0.2	-12.9
79-03-R	OIL TEMPERATURE SENSOR - S2335-1	0750635	0.2	-6.4

Figure 6-9 (Sheet 6)